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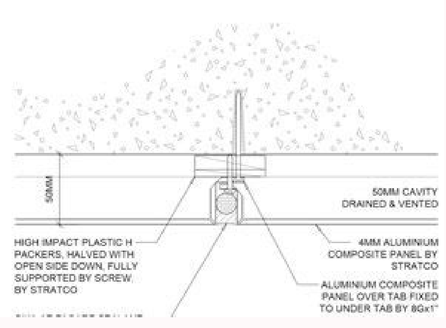

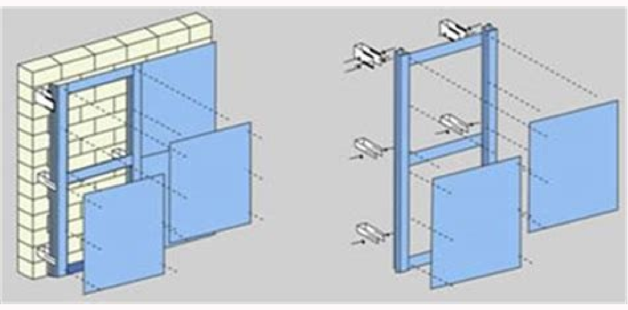

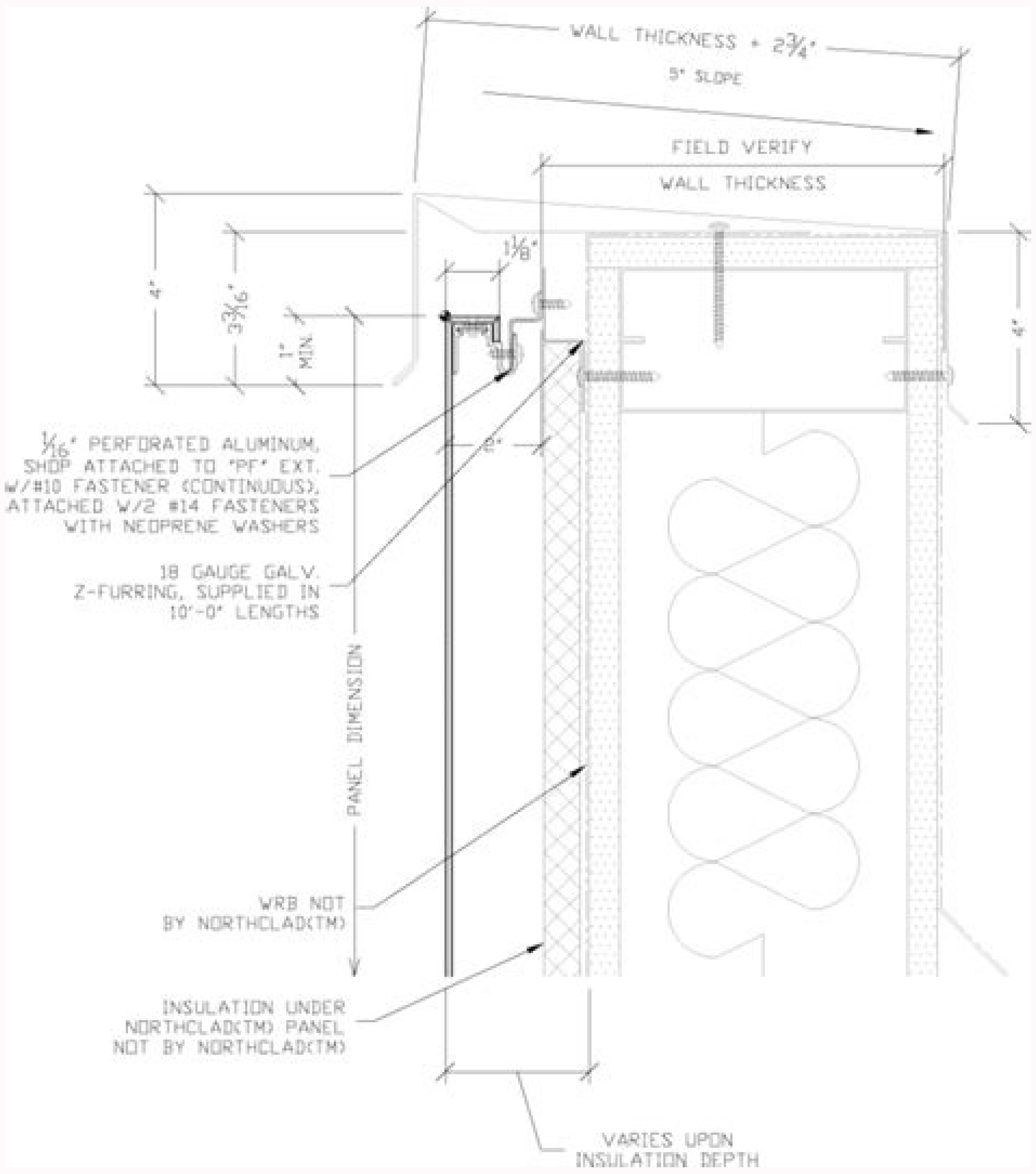

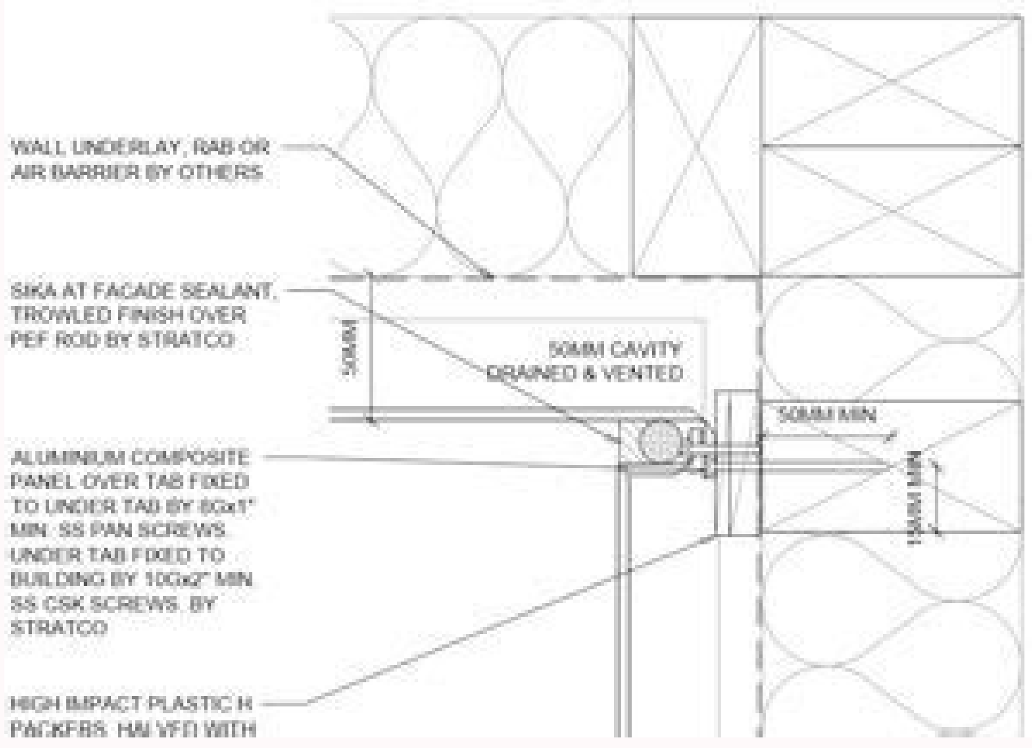

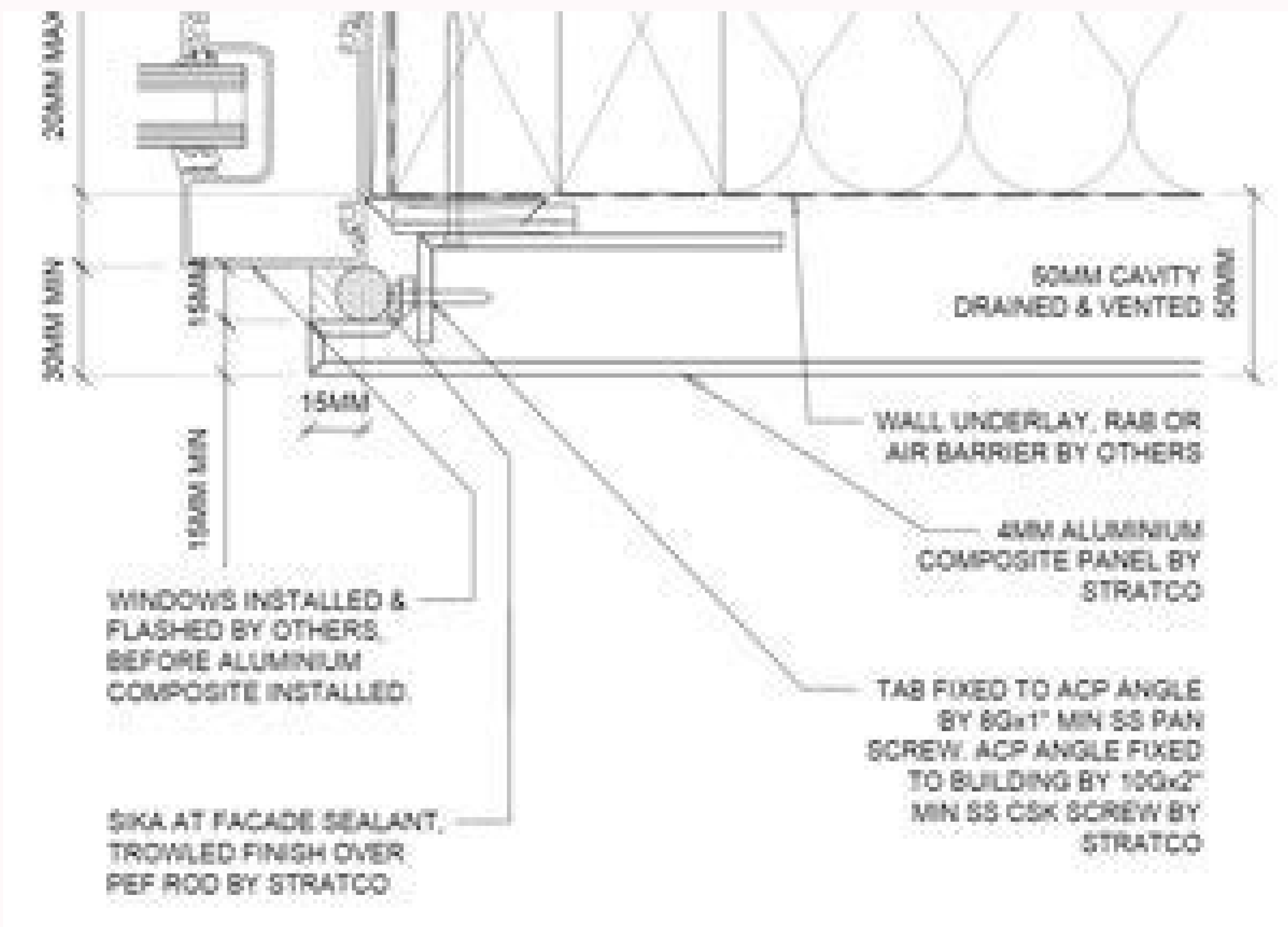

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CPH Group Renda Helin Design & Interiors created these officespaces for manufacturing company Emre Group, located in Istanbul, Turkey. Details of the Lego Movie and nods to construction inspired the design which features a pallette of vibrant yellow and monochromatic black and greys against a white backdrop. Floor, wall and window graphics are used extensively to reinforce the construction theme, even concrete walls proved to be no barrier. Gallery > Inspiration Studco Top Hats are suitable for use with a wide variety of lining materials such as fibre cement sheet, aluminium composite panels, AAC panels and timber feature linings, and can be installed in both a horizontal and vertical orientation depending on the needs of the structure. Studco top hats may also be used for internal applications where heavier loads are used or greater design pressures are specified. 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Access through your institutionVolume 284, 15 March 2022, 115120 rights and contentThere is an ever-pressing need within engineering to reduce structural mass and, hence, improve structural efficiency. From a design viewpoint, two options are available to improve structural efficiency: change the material used or change the geometric configuration [1]. Lattice structures are highly efficient geometric configurations formed of a network of individual elements connected at joints. By grouping material together into localised, discrete elements (or members), they can move material far away from the axis of bending or torsion. Further efficiency gains are achieved as the elements that make up the structures carry primarily axial loads, meaning material within each element is loaded equally [2].Advanced composite materials are also often selected for their ability to maximise structural efficiency. A key feature of these materials is that their anisotropic behaviour can be tailored to maximise material properties in a given direction. Since their discovery, the use of composites has been overwhelmingly focused on monolithic plate and solid section structures. Implementing composites within lattice structures presents a major advantage: the material anisotropy can be tailored to maximise properties along the primarily axially loaded elements. Composite lattice structures therefore present a synergistic combination of material and geometry, in which the geometry directs the forces to align with the material's primary fibre direction. The result is incredibly high structural efficiencies.While composite lattice structures have seen growing research interest in recent years, their use in industrial applications has, so far, been limited. A possible reason for this is the notorious difficulties associated with joining composite parts. Manufacturing of lattice structures traditionally involves the assembly and joining of many parts, making the use of composites less attractive.Broadly speaking, there are two approaches to manufacturing composite lattice structures. In the first, more traditional approach, the lattice elements are premanufactured and then later assembled to form the lattice geometry. This approach will always result in a discontinuity of the fibres at the joints between adjacent elements. The second approach uses more novel manufacturing techniques to form the lattice elements and the geometry at the same time. The latter approach results in some continuity of fibres between adjacent members and can greatly reduce or remove the need to join many composite parts. This approach, however, presents its own challenges. For example, forming composite materials in an open lattice geometry makes consolidation of the members difficult. Additionally, creating sufficient joint strength between members that do not share continuous fibres is still a challenge.Within the literature, there are a range of technologies that could be classed as composite lattice structures. Most often, the definition of the technologies is more dependent on the manufacturing process than the final structural configuration itself. How these structures can be effectively manufactured is the key question to unlocking their potential. This paper will therefore review composite lattice structures with a focus on manufacturing.The majority of composite lattice structures can be grouped into the following three categories:•Composite lattice beam (CLB) structures•Composite grid stiffened panel (CGSP) structures•Composite lattice core sandwich panel (CLCSP) structuresComposite lattice beams are three-dimensional (3D) arrangements of composite members that macroscopically form one-dimensional beam-like structures. Such structures are highly efficient in flexure due to their ability to move aligned fibre material far from the beam neutral axis. This efficiency in bending, coupled with high axial stiffness and strength, also makes them attractive solutions as lightweight compression columns. CLBs are typically formed of longitudinal (or chord) members that run parallel to the beam axis, and shear members that run diagonally to the beam axis. Note, that shear members are sometimes formed via helical winding and hence are often called 'helical members' or 'helicals'. Within a CLB, the longitudinal members predominately carry the axial and bending loads while the shear members carry the shear and torsional loads.Composite grid stiffened panels are structures formed from a series of composite stiffeners (often referred to as ribs) that are arranged into a repeating two-dimensional (2D) planar grid pattern. The grid structure is normally bonded, or co-cured, to a composite skin to give a continuous surface. The planar grid is often curved, giving a cylindrical or conical surface. Such grid structures provide the oldest examples of composite lattices and will be discussed in Section 3. CGSPs show excellent resistance to axial compression loading and as such have been used in launch vehicle adaptors and payload connectors, such as the Russian Proton-M rocket [3]. While most examples use a single skin, the use of two skins to form grid stiffened sandwich panels has also been studied.Replacing conventional honeycomb or foam sandwich panel cores with 3D lattices formed of composite struts is an idea currently being explored by several research groups. This category of structures is referred to hereon as composite lattice core sandwich panels and will be discussed in Section 4. Similar to CLBs, CLCSP structures are also able to move material far from the neutral axis with little mass penalty, and thus offer high levels of structural efficiency compared to conventional core materials.Composite lattice technologies that do not fall into any of these three categories normally involve manufacturing processes that can be used to produce more general 3D lattice geometries. A brief discussion regarding these structures will be given in Section 5; however much greater focus will be given to technologies of the three main categories. It is also worth noting that several deployable space structures could be classed as composite lattice structures, however this review will focus on static, non-mechanised structures.This section will introduce and discuss the different families of technologies used to manufacture composite lattice beams.Reinforcement members for structural panels typically take the form of stiffeners (or ribs), running in two or more directions, often in a repeating grid or lattice, offering improved load resistance for a modest mass increase. This method of reinforcement has been used in engineering for decades where, as early as the 1960's, optimisation of stiffened panels was attempted through varying stiffener size, thickness and placement, to meet load requirements while minimising weight [41]. AsA large body of research has been undertaken into the use of 3D lattice structures as sandwich panel cores. Before considering this third classification of composite lattices however, it is worth noting that while composite grids such as kagome-grid [61], [69], [70], isogrids, and anisogrids, have also been used for sandwich core design. Still, they are considered as 2D lattice structures in which the elements are mostly deformed in shear and bending when loaded out of plane. These conceptsA small number of existing composite lattice concepts do not fit into the previously discussed categories of beams, grids, and sandwich panels. These examples tend to involve manufacturing processes capable of producing more general 3D composite lattices that are not limited to a specific macroscopic form.Similar to the other lattice categories, the methodologies reviewed here focus on two main manufacturing philosophies: single-component or multi-component assembled lattices. ManufacturingThis paper has presented, for the first time, a review of all existing literature in the field of composite lattice structures. The technologies discussed offer an alternative to the paradigm of monolithic, laminated composite structures. As has been demonstrated in many studies, the key benefit of composite lattices is their incredibly high structural efficiencies. However, key manufacturing challenges remain that have, so far, limited their adoption within industry.The majority of compositeThe authors declare that they have no known competing financial interests or personal relationships that could have appeared to influence the work reported in this paper.This work was supported by the Engineering and Physical Sciences Research Council through the EPSRC Centre for Doctoral Training in Advanced Composites for Innovation and Science [grant number EP/G036772/1].This is a review article, and therefore all data underlying this study is cited in the references.Y. 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Sui et al.The 3D printing process for continuous fibers offers a new approach to the fabrication of lattice structures. In this paper, a planning method for a single cell trajectory of the lattice structure is proposed. Lattice sandwich structures with different strut numbers and angles are prepared according to this method. Out-of-plane compression tests are also carried out. Theoretical prediction models for the compressive strength of tetrahedral, pyramidal and hexagonal cone structures are developed, with errors within 32.8% compared to experimental results. The effects of strut angle, number and material on the compressive properties and energy absorption capacity of the lattice sandwich structure are analyzed. The pyramidal structure with continuous fibers reinforced 60° struts has the optimum specific compression strength of 10.115MPa/(g/cm3). Defects such as voids, fiber pull-out and cracking of the resin matrix are observed in the strut section. It is found that strut fracture and delamination are the main failure modes of lattice structures under compressive loading.In this work, a new type of energy-absorbing thin-walled tubes with concave angles is proposed by a unique structural design method to improve the crashworthiness performance of the traditional hexagonal thin-walled tube (TH). These concave tube structures (CTSs) are named CTS1, CTS2, and CTS3 respectively, while CTS1 is developed by a common design method. The crushing behaviors of the CTS3 are investigated by quasi-static compression experiments and numerical simulations. The crashworthiness and energy dissipation mechanism of all the tubes are investigated, and the results demonstrate that the CTS3 exhibits superior energy absorption capability than the other proposed tubes and TH with the same mass. Then, the mean crush resistance of the CTSs is predicted by theoretical analysis, and the influences of slenderness rate, boundary condition, and loading rate on the crushing responses of CTS3 are performed by numerical analysis. Besides, the comparative analysis of performances of the CTS3 and the typical concave tubes (TCTs) is carried out, and the results indicate that the CTS3 has the best energy absorption capacity among these tubes. In addition, the optimal structure parameters of CTS3 are explored to enhance the capacity of energy absorption further.The stacking sequence has a great influence on the performance of composite structures especially when it comes to stress concentration effects. In the current research, various orientation angles and layups for a composite plate with a central hole are investigated and the optimized layup is determined for the desired thickness. The objective of the optimization is the minimization of the interlaminar peeling stress around the hole. Failure strength around the hole is the basis of the constraint incorporating Hashin's failure criteria for each lamina individually. For this purpose a two-step analysis is carried out: Primarily a finite element analysis is performed using ABAQUS to determine the constraint for orientation angles. Then, the pattern search method is utilized to obtain the optimized layup using the interaction of MATLAB and ABAQUS software. Moreover, a sensitivity analysis is carried out to determine the effect of the material properties, the size of the hole and the initial guess applied for the pattern search solution upon the optimal layup sequence.Two-dimensional star-shaped honeycombs (2D SSHs) exhibit an effective negative Poisson's ratio due to the abundant internal space and re-entrant angle, while the strength and the deformation of the three-dimensional (3D) SSHs are still limited owing to the finite tuneability of the tip angle. By adding different tip re-entrant angles into the SSH, multiple improved star-shaped honeycombs (ISSHs) with tunable Poisson's ratio are proposed. The in-plane elastic properties, including the effective Young's moduli and effective Poisson's ratio, are both derived by 2D analytical model using the energy method. The finite element simulation and compression experiment are used to verify the correctness of theoretical results. Based on the work, the deformation mechanism of the 3D ISSHs is discussed by quasi-static compression experiment and numerical simulation. The simulation results and experimental results show a great agreement with theoretical results. Different tip angles heighten the normalized Young's modulus and make the Poisson's ratio more tunable, respectively. In addition, the 3D ISSHs show an enhanced effect among higher strength and stability while bearing the compression load. This work provides a good reference for constructing 3D symmetrical multicellular structures, especially honeycombs.The goal of this research is to evaluate the variability of the buckling load in thin orthotropic cylindrical shells due to the deviation with respect to the nominal values, for both the geometrical and material properties. In this research also the fiber orientation i.e the alignment with respect to the cylinder axis was considered. The research is developed by comparing the results obtained through analytical formulas and numerical results (f.e.m.) for isotropic and orthotropic materials. The coefficient of variation for critical buckling load (for symmetrical and non-symmetrical deformation), assumed for comparing the standard deviation values, is linear with the variation of material and geometrical parameters. The standard deviation on critical buckling load considering both the material and geometrical variations and fiber orientation can be very well described by a surface. From these surfaces it is important to underline that to quantify the dispersion values on the critical buckling load, the material properties assume the fundamental role with respect to the fiber orientation. The main implications of these results are correlated to the fact that, for example, during the manufacturing process, the geometrical and material variability is more important than the fiber orientation in order to have a reliability component.The composite sandwich plate with hourglass lattice cores (CSP-HLC) is a novel cellular structure that can increase the width-to-length ratio and reduce the inter-node spacing. However, its static and dynamic behaviors are difficult to analyze due to the complex microstructures. In this work, a computational homogenization method is proposed for calculating the constitutive parameters. A reduced-order plate model is then derived through dimensional reduction from the three-dimensional orthotropic thermoelasticity framework. The effectiveness and accuracy of the proposed model were verified by comparing with the static-displacement and free-vibration results of 3D direct numerical simulations. The parameter analysis showed that the different material and structural parameters of the hourglass lattice core had different effects on the equivalent stiffnesses and natural frequencies of the CSP-HLC. Compared with the composite sandwich plate with pyramidal lattice cores, the displacement of the CSP-HLC was smaller under the same load and boundary conditions, and the natural frequencies were also smaller, which may have been because the additional equivalent density of the CSP-HLC influenced the frequency more predominantly than the extra equivalent stiffness.Thermoplastic composites offer many opportunities for aeronautics. Among them is the possibility of overmolding composite parts to obtain finished "netshape" parts in medium series. This paper presents the complexity of the manufacturing method and the manufacturing defects that may occur in some grid-stiffened overmolded parts using aeronautic thermoplastics. The results of impact tests up to 15 J carried out at various locations of the plates and the grid show that, due to manufacturing defects at the overmolded grid/plate interface, very significant delaminations can be observed.View full text

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